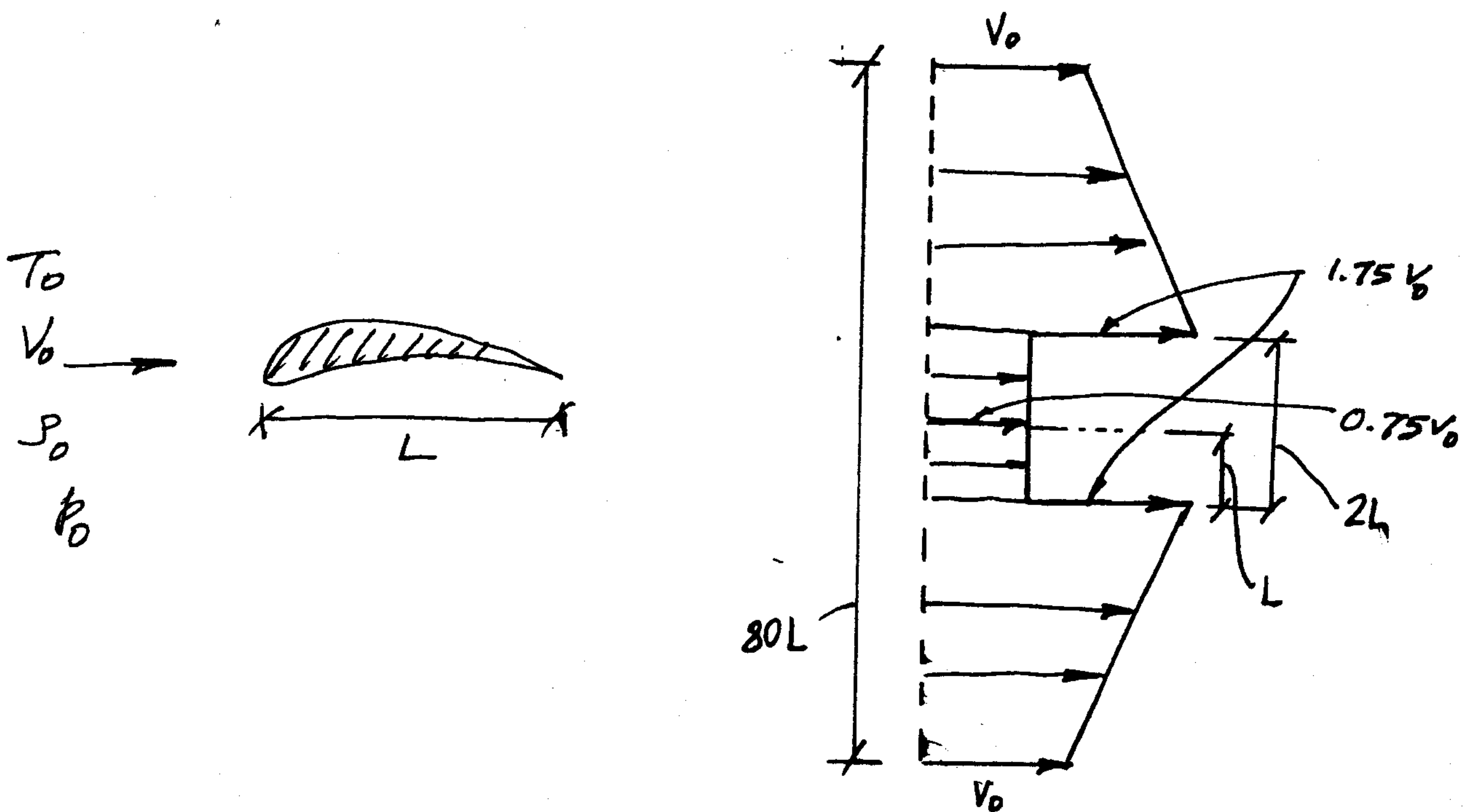


CONSERVATION OF LINEAR MOMENTUM (INTEGRAL FORM)

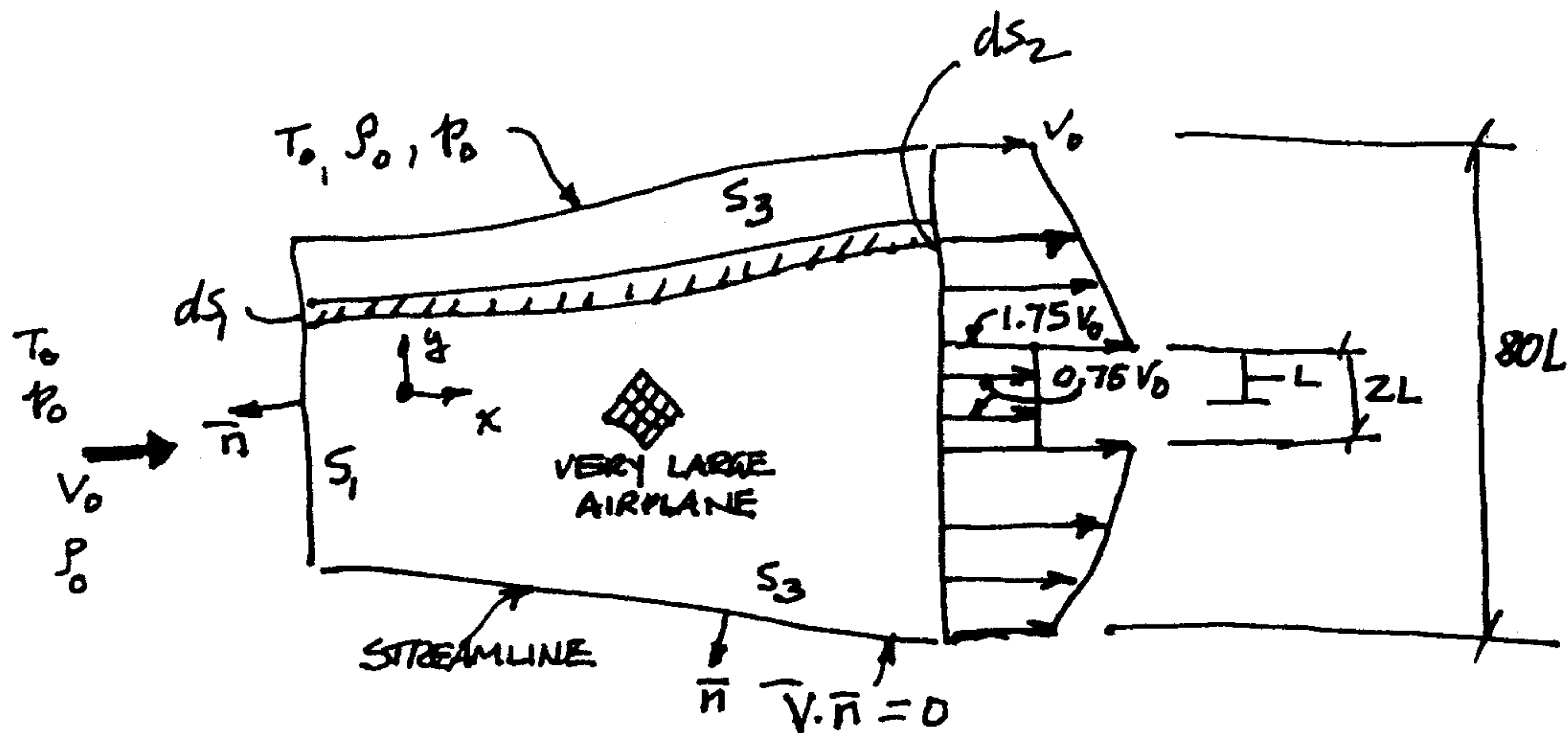
DRAG CALCULATION

AN INTERNATIONAL CONSORTIUM OF AIRCRAFT COMPANIES HAS AGREED TO BUILD A LOW SPEED VERY LARGE AIRPLANE (VLA). THE VLA CAN TRANSPORT A MAXIMUM OF 800 PASSENGERS A DISTANCE OF 5,000 NAUTICAL MILES. THE WAKE PROFILE (VELOCITY PROFILE) PRODUCED BY A TWO-DIMENSIONAL AIRFOIL MODEL OF A WING SECTION OF THE VLA HAS BEEN SHOWN TO BE:



COMPUTE/CALCULATE AND DISCUSS FULLY THE NET FORCE ACTING ON THE AIRCRAFT AIRFOIL IN THE DIRECTION OF THE FREE STREAM VELOCITY.

SOLUTION



ASSUMPTIONS:

1. STEADY FLOW
2. INCOMPRESSIBLE FLUID
3. VERY SMALL VISCOUS EFFECTS
4. UNBOUNDED FLUID
5. UNIFORM UPSTREAM CONDITIONS: T_0, p_0, V_0, ρ_0
6. 2-D FLOW
7. ON S_3 : $\vec{v} \cdot \vec{n} = 0, p = p_0, \rho = \rho_0$
8. ON S_2 : $\vec{v}_2 = v_2(y) \vec{i}, \rho = \rho_0, p = p_0$
9. ZERO GRAVITY EFFECTS
10. WAKE PROFILE IS SYMMETRIC W.R.T. THE X-AXIS

FROM CLASS NOTES:

$$\text{DRAG} = - \iint_{\text{SURF}} (p - p_0) \vec{n} \cdot \vec{i} \, d\text{SURF} - \iint_{\text{SURF}} \rho \vec{v} \cdot \vec{i} (\vec{v} \cdot \vec{n}) \, d\text{SURF}$$

FROM ASSUMPTIONS 5, 7, AND 8: $\iint_{\text{SURF}} (p - p_0) \vec{n} \cdot \vec{i} \, d\text{SURF} = 0$

$$\therefore \text{DRAG} = - \iint_{\text{SURF}} \rho \vec{v} \cdot \vec{i} (\vec{v} \cdot \vec{n}) \, d\text{SURF} = \iint_{S_1} \rho_0 v_0^2 \, dS_1 - \iint_{S_2} \rho_0 v_0^2 (y) \, dS_2$$

APPLY CONSERVATION OF MASS PRINCIPLE TO ARBITRARY STRIP BETWEEN TWO STREAMLINES CONNECTING S_1 AND S_2 , AND OBTAIN:

$$\rho_0 V_0 dS_1 = \rho_0 V_2(y) dS_2$$

SUBSTITUTE AND CONTINUE AS:

$$\text{DRAG} = \iint_{S_2} \rho_0 V_0 V_2(y) dS_2 - \iint_{S_2} \rho_0 V_0^2 dS_2 = \iint_{S_2} \rho_0 V_2(y) (V_0 - V_2(y)) dS_2$$

USING ASSUMPTION 10:

$$\text{DRAG} = 2 \int_0^{40L} \rho_0 V_2(y) (V_0 - V_2(y)) dy$$

PROM WAKE PROFILE

$$V_2(y) = \begin{cases} 0.75V_0, & 0 < y < L \\ 1.75V_0 + m^* (L-y), & L < y < 40L \end{cases} \quad m^* \equiv \frac{V_0}{52.0L}$$

SUBSTITUTE AND EVALUATE INTEGRALS:

$$\text{DRAG} = -197.25 \rho_0 V_0^2 L$$

THIS VERY LARGE AIRPLANE PRODUCES A NET THRUST!!