SUMMARY
THIN AIRFOIL THEORY

PARAMETER	SYMMETRIC AIRFOIL	CAMBERED AIRFOIL
LIFT COEFFICIENT, C_ℓ	$2\pi\alpha$	$2\pi\left[\alpha + \frac{1}{\pi}\int_0^\pi \left(\frac{dy}{dx}\right)_\theta^{MCL} (\cos\theta - 1)\, d\theta\right]$
C_ℓ SLOPE VS. α	2π	2π
CENTER OF PRESSURE, X_{cp}	$c/4$	$c/4 - \frac{\pi c}{4}\frac{A_2 - A_1}{C_\ell}$
$C_{m_{LE}}$	$-\frac{\pi\alpha}{2} = -\frac{C_\ell}{4}$	$-C_\ell/4 + \frac{1}{4}\pi(A_2 - A_1)$
X_{ac}	$c/4$	$c/4$
$C_{m_{ac}}$	0	$\frac{1}{2}\int_0^\pi \left(\frac{dy}{dx}\right)_\theta^{MCL} (\cos 2\theta - \cos\theta)\, d\theta$
ANGLE OF ZERO LIFT, α_0	0	$-\frac{1}{\pi}\int_0^\pi \left(\frac{dy}{dx}\right)_\theta^{MCL} (\cos\theta - 1)\, d\theta$

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