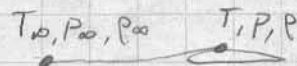


a) Isentropic relation between " ∞ " and airfoil point



$$\frac{T}{T_\infty} = \left(\frac{P}{P_\infty}\right)^{\frac{\gamma-1}{\gamma}} = \left(\frac{3.6 \times 10^4}{4.35 \times 10^6}\right)^{\frac{1}{3.5}} = 0.94737 \rightarrow T = 0.94737 T_\infty = 232.1 \text{ K}$$

$$\text{State eqn: } \rho = \frac{P}{RT} = \frac{3.6 \times 10^4 \text{ Pa}}{287 \text{ J/kgK} \cdot 232.1 \text{ K}} = 0.540 \text{ kg/m}^3$$

$$\text{b) } \frac{T}{T_0} = \left(\frac{P}{P_0}\right)^{\frac{\gamma-1}{\gamma}} = \left(\frac{1 \text{ atm}}{10 \text{ atm}}\right)^{\frac{1}{3.5}} = 0.518 \rightarrow T = 0.518 \cdot 500 \text{ K} = 259.0 \text{ K}$$

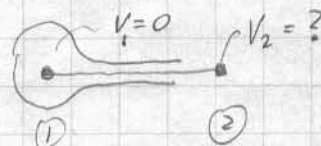
$$\rho = 1 \text{ atm} = 1.013 \times 10^5 \text{ Pa}, \quad \rho = \frac{P}{RT} = \frac{1.013 \times 10^5 \text{ Pa}}{287 \text{ J/kgK} \cdot 259 \text{ K}} = 1.363 \text{ kg/m}^3$$

c) Convert to SI units: $V = 1300 \text{ ft/s} = \frac{1300 \text{ ft/s}}{3.28 \text{ ft/m}} = 396.34 \text{ m/s}$, $T = 480 \text{ R} = \frac{480 \text{ R}}{9/5 \text{ R/K}} = 266.7 \text{ K}$

$$c_p = 1003 \text{ J/kgK}: \quad h_0 = c_p T + \frac{1}{2} V^2 = 1003 \cdot 266.7 + \frac{1}{2} (396.34)^2 = 346009 \text{ m}^2/\text{s}^2$$

$$\text{or } h_0 = 346000 \text{ m}^2/\text{s}^2 \cdot (3.28 \text{ ft/m})^2 = 3.723 \times 10^6 \text{ ft}^2/\text{s}^2$$

d) Adiabatic flow: $h_0 = h$ or $h_01 = h_02$



$$h_1 + \frac{1}{2} V_1^2 = h_2 + \frac{1}{2} V_2^2, \quad h_1 = c_p T_1 = 1003 \text{ J/kgK} \cdot 1000 \text{ K} = 1.003 \times 10^6 \text{ m}^2/\text{s}^2$$

$$h_2 = c_p T_2 = 1003 \text{ J/kgK} \cdot 600 \text{ K} = 0.602 \times 10^6 \text{ m}^2/\text{s}^2$$

$$V_2 = \sqrt{2(h_1 - h_2)} = 895.8 \text{ m/s}$$

e) Isentropic (and adiabatic) flow between " ∞ " and airfoil point



$$P_\infty = 0.61 \text{ atm} = 0.61 \text{ atm} \cdot 1.013 \times 10^5 \text{ Pa/atm} = 0.618 \times 10^5 \text{ Pa}, \quad T_\infty = \frac{P_\infty}{\rho_\infty R} = \frac{0.618 \times 10^5}{0.819 \cdot 287} = 262.9 \text{ K}$$

$$\frac{T}{T_\infty} = \left(\frac{P}{P_\infty}\right)^{\frac{\gamma-1}{\gamma}} = \left(\frac{0.50 \text{ atm}}{0.61 \text{ atm}}\right)^{\frac{1}{3.5}} = 0.9448 \rightarrow T = 0.9448 \cdot T_\infty = 248.4 \text{ K}$$

$$\text{Adiabatic: } h_0 = h_\infty \rightarrow c_p T + \frac{1}{2} V^2 = c_p T_\infty + \frac{1}{2} V_\infty^2 \rightarrow V = \sqrt{2(c_p T_\infty + \frac{1}{2} V_\infty^2 - c_p T)} = 345.1 \text{ m/s}$$

$$\text{f) } \frac{T}{T_\infty} = \left(\frac{P}{P_\infty}\right)^{\frac{\gamma-1}{\gamma}} = \left(\frac{0.30 \text{ atm}}{0.61 \text{ atm}}\right)^{\frac{1}{3.5}} = 0.8165 \rightarrow T = 0.8165 T_\infty = 214.7 \text{ K}$$

$$V = \sqrt{2(c_p T_\infty + \frac{1}{2} V_\infty^2 - c_p T)} = 432.2 \text{ m/s}$$